



## Simulation of Precise and Safe Landing near a Plume Source on Enceladus

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# **Motivation - Enceladus Lander mission**

Shielded Electronics

- **ISTA** involved in EnEx-1 project (2012-2015) for **Enceladus Lander mission design**
- Critical technology identified for landing mission: Advanced landing GN&C

condensation and release of latent heat

> vapor + salty liquid droplets

> > top of

water table

Porco et al., 2014

geyser (vapor
+ ice particles)



K. Konstantinidis et al., A lander mission to probe subglacial water on Saturn's moon Enceladus for life, Acta Astronaut. 106 (2015) 63–89

Low Gain Antenna





# Landing on the south pole of Enceladus

et al., 2010

### Topography

- "V" shaped canyons, 100–150 m high, 200–250 m deep
- Bottom: 50–100 m wide
- floor is interspersed with obstacles

#### **Terrain texture**

- plume fallout deposit layer up to 10s of meters near plumes
- Also exposed icy crust

### **Thermal environment**

Concentrated hotspots around plume sources (~10 m)

## Albedo of Enceladus: 0.99

### **Polar region**

- low sun elevation angles
- canyon floors often in darkness

## **Planetary protection**

possible direct access to ocean, PP more strict than e.g. Mars special regions







# Lander GN&C operations



## Lander GN&C system



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# **Current research - GN&C for Enceladus Landing**



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## Terrain and sensor simulation





**2**. Delaunay triangulation



3. Terrain illumination







# **TRN-aided inertial navigation**

1. Lander nav. state J. Sola, 2007  $\mathbf{\hat{x}}_k = \begin{bmatrix} \text{position} \\ \text{velocity} \end{bmatrix}$  $\mathbf{P}_k = egin{bmatrix} \Sigma_{pp} & \Sigma_{pv} \ \Sigma_{vp} & \Sigma_{vv} \end{bmatrix}$ 2 3 **2**. Prediction step  $\hat{\mathbf{x}}_k = \mathbf{F}_k \hat{\mathbf{x}}_{k-1} + \mathbf{B}_k \mathbf{u}_k$ Sensor observations  $\mathbf{P}_{k} = \mathbf{F}_{k} \mathbf{P}_{k-1} \mathbf{F}_{k}^{T} + \mathbf{Q}_{k}$ Control -Control + noise forces meas. + noise 3. Correction step  $\hat{\mathbf{x}}_{k}^{\prime} = \hat{\mathbf{x}}_{k} + \mathbf{K}^{\prime} (\overrightarrow{\mathbf{z}_{k}} - \mathbf{H}_{k} \hat{\mathbf{x}}_{k})$ Prediction Correction Nav. state + **IMU** motion equations covariances step step  $\mathbf{P}'_k = \mathbf{P}_k - \mathbf{K}' \mathbf{H}_k \mathbf{P}_k$ Extended  $\mathbf{K}' = \mathbf{P}_k \mathbf{H}_k^T (\mathbf{H}_k \mathbf{P}_k \mathbf{H}_k^T + \mathbf{R}_k)^{-1}$ **Kalman Filter** bzarg.com



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## **Sensor observations**

**Camera: Simultaneous Localization and Mapping (SLAM)** 



## Hazard Detection and Avoidance – Fuzzy reasoning







## Hazard Detection and Avoidance – Landing site evaluation



# **Guidance – Convex guidance (G-FOLD)**

(3)

(4)

Problem 1 Non-Convex Minimum Fuel Planetary Landing Problem

$$\begin{aligned} \max_{t_f, \mathbf{T}_c} m(t_f) & \text{subject to:} \\ \dot{\mathbf{x}}(t) &= A\mathbf{x}(t) + B\left(\mathbf{g} + \frac{\mathbf{T}_c(t)}{m}\right) \\ \dot{m}(t) &= -\alpha \|\mathbf{T}_c(t)\| \\ \mathbf{x}(t) &\in \mathbf{X} \quad \forall t \in [0, t_f], \\ 0 &< \rho_1 \leq \|\mathbf{T}_c(t)\| \leq \rho_2, \quad \hat{\mathbf{n}}^T \mathbf{T}_c(t) \geq \|\mathbf{T}_c(t)\| \cos \theta, \\ m(0) &= m_0 \\ \mathbf{r}(0) &= \mathbf{r}_0, \quad \dot{\mathbf{r}}(0) &= \dot{\mathbf{r}}_0, \\ \mathbf{r}(t_f) &= 0, \quad \dot{\mathbf{r}}(t_f) &= \mathbf{0}. \end{aligned}$$

Acikmese et al.,

SPACE TECHN



- **Optimization objective:** minimize propellant used
- Dynamics
  - Constraints
- (5) Initial and final conditions
   (6)
- (7) (8) **Convex Optimization:** G-FOLD
- (9) algorithm



## Landing simulation tool structure



# **Visual navigation simulation**







## Hazard Detection and Avoidance simulation



## **Guidance Simulation**

### **Demonstation of G-FOLD capabilities**



### Modified reachability ellipse verification



# Example challenging trajectory and thrust arc





# Summary and future work

## What we did so far

- Created *tool to simulate landing* on Enceladus
- Incorporating all functions necessary:
  - Terrain Relative Navigation
  - Hazard Detection and Avoidance
  - Guidance
  - Realistic sensor input simulation

## Work remaining to be done

- Finalize tool development
- Monte Carlo simulations and sensitivity analyses to *verify fulfilment of landing requirements*
- Focus on reliability: apply fault management methods and simulate deduced fault scenarios
   Future work
- Test landing on quadrocopter testbed under development in ISTA





# Thank you! Questions?

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